

AD-A097 194

AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)  
COARSELY RANDOM CRACKING IN ONE-CRACK FATIGUE MODELS.(U)  
MAR 80 D 6 FORD  
ARL/STRUC-382

F/6 20/11

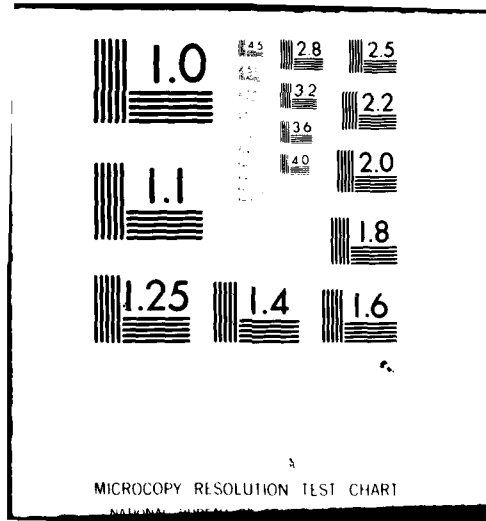
UNCLASSIFIED

NL

For  
AD-A097 194



END  
DATE  
FILMED  
5-81  
DTIC





**DEPARTMENT OF DEFENCE**  
**DEFENCE SCIENCE AND TECHNOLOGY ORGANISATION**  
**AERONAUTICAL RESEARCH LABORATORIES**  
**MELBOURNE, VICTORIA**

STRUCTURES REPORT 382

**COARSELY RANDOM CRACKING IN ONE-CRACK  
FATIGUE MODELS**

THE UNITED STATES NATIONAL  
TECHNICAL INFORMATION SERVICE  
IS AUTHORISED TO  
REPRODUCE AND SELL THIS REPORT

by

D. G. FORD

Approved for Public Release.



DTIC  
SELECTED  
APR 2 1981

A

© COMMONWEALTH OF AUSTRALIA 1980

COPY No 14

MARCH 1980

81 4 2 146

AD A 097 194

DTIC FILE COPY

DEPARTMENT OF DEFENCE  
DEFENCE SCIENCE AND TECHNOLOGY ORGANISATION  
AERONAUTICAL RESEARCH LABORATORIES

14 ARL/STP.11-27

STRUCTURES REPORT 382

**COARSELY RANDOM CRACKING IN ONE-CRACK  
FATIGUE MODELS.**

by

D. G. FORD

**SUMMARY**

The continuity equation has been applied to fatigue crack models in which all cracks begin together to form a family of smooth trajectories proceeding at different rates. When the rates at a particular crack length are log-Normally distributed, it is possible to estimate the effect of the mean and variance of crack life in a reliability situation. For a logarithmic variance 0.1 of crack rates, the standard deviation of life is reduced by 14% approximately.

This crack model has been applied to a previous one-crack analysis to allow randomness in both initiation and cracking.

## DOCUMENT CONTROL DATA SHEET

Security classification of this page: Unclassified

1. Document Numbers

- (a) AR Number:  
AR 001 810
- (b) Document Series and Number:  
Structures Report 382
- (c) Report Number:  
ARL Struc. Report 382

2. Security Classification

- (a) Complete document:  
Unclassified
- (b) Title in isolation:  
Unclassified
- (c) Summary in isolation:  
Unclassified

3. Title: COARSELY RANDOM CRACKING IN ONE-CRACK FATIGUE MODELS

4. Personal Author(s):

D. G. Ford

5. Document Date:

Published - March, 1980  
Work - January, 1979

6. Type of Report and Period Covered:

7. Corporate Author(s):

Aeronautical Research Laboratories

8. Reference Numbers

- (a) Task: AIR 76 156
- (b) Sponsoring Agency:

9. Cost Code:

27 7030

10. Imprint:

Aeronautical Research Laboratories,  
Melbourne

11. Computer Program(s)

(Title(s) and language(s)):

12. Release Limitations (of the document): Approved for public release

12-0. Overseas:

N.O.	P.R.	I	A	B	C	D	E
------	------	---	---	---	---	---	---

13. Announcement Limitations (of the information on this page): No limitation

14. Descriptors:

Fatigue (materials)      Crack propagation  
Cracking (fracturing)      Crack initiation  
Mathematical models

15. Cosati Codes:

1113  
2012

16.

ABSTRACT

*The continuity equation has been applied to a fatigue crack model in which all cracks begin together to form a family of smooth trajectories proceeding at different rates. When the rates at a particular crack length are log-Normally distributed, it is possible to estimate the effect on the mean and variance of crack life in a reliability situation. For a logarithmic variance 0.1 of crack rates, the standard deviation of life is reduced by 14%, approximately.*

*This model has been applied to a previous one-crack analysis to allow randomness in both initiation and cracking.*

## CONTENTS

	Page No.
NOTATION	
1. INTRODUCTION	1
2. DISTRIBUTION OF CRACKING LIFE	1
2.1 Nature of Random Cracking	1
2.2 Rate Divergence	2
2.3 Crack Length Density	3
2.3.1 Boundary Conditions	3
2.4 Density of Crack Life	4
3. RUNAWAY CRACKS	5
3.1 Reduction of Life Range	5
3.2 Telescoping	5
4. LIFE DISTRIBUTION	6
4.1 Moment Generating Function of Crack Time	7
4.2 Crack Life MGF for Small Variance	8
4.3 Heuristic Approximations	9
4.3.1 Example	9
5. CONCLUSIONS	10
REFERENCES	
DISTRIBUTION	

A

## NOTATION

$A t$	Random crack length conditional upon time $t$
$a, (a t)$	Crack length (at time $t$ )
$a_0 = a t_0$	Length of initial cracks
$f = f(a t)$	Crack length density
$f$	Probability density function according to subscript
$g, h$	Heuristic constants, Section 4.3
$G, G(t_a)$	Defined by Equation (4.5)
$H, H(t_a)$	$= E(t_a) - ut_a$
$M_a$	Moment generating function of median cracking life
$M_A(u)$	Moment generating function of crack life
$M_t(u)$	MGF of initiation time
$M_\phi(u)$	MGF of total life
$r(a) = r(t_a)$	Risk or hazard rate
$r_0 = r(0)$	Hijack risk
$R, R(a, t)$	Generic cracking rate
$R_0(a) = R_0(t_a)$	Median growth rate
$R_0 = R_0(0)$	Initial median growth rate
$t$	Overall time or life
$t_a$	Median crack growth time
$t_A$	$t_a$ with random cracking
$t_0 = t - t_a$	Initiation
$T_a$	Median life to runaway cracking
$u$	Transform variable for MGFs
$X$	Random factor for particular cracks
$x$	Particular value of $X$
$\phi(t)$	Density of life
$\phi_A(t_A)$	Density of crack time with random rates
$\mu_{ra}$	$r$ th moment of median crack life
$\xi$	$-\ln x$
$\sigma^2$	Variance of $x$
$\sigma_a^2$	Variance of median crack life

## 1. INTRODUCTION

Until now, the general theory of structural fatigue<sup>1,2</sup> has treated the growth of cracks in structures with two distinct stages—viz. crack initiation or damage, followed by crack growth. The idealisation implied in those papers is that variability is introduced by random initiation time, while structural interactions are related to the growth of cracks. In reality, the second assumption is fairly accurate but there is randomness in the crack growth exhibited by nominally identical structures. So far, allowing for this has defied analysis for multiply-cracked structures; the present report extends the previous work to allow random cracking rates in single-crack models.

It begins with an exact treatment of the crack growth models of Payne<sup>3</sup> or Hooke<sup>4</sup> which is then combined with random initial lives. Previous results for life and moment generating functions are generalised and there is some discussion of the low order effects of random crack rates.

## 2. DISTRIBUTION OF CRACKING LIFE

This replaces the reliability type of life density<sup>1</sup> previously found for one-crack models with deterministic cracking, and indeed it will transpire that the present result is obtainable from that model by simple substitution. Before beginning, however, the randomness in crack rate must be described more closely.

### 2.1 Nature of Random Cracking

For the earlier deterministic cracks,<sup>1</sup> the possible trajectories were a group of curves obtainable from one another by translation. These were indexed by the initiation time  $t_0$ , but all satisfied the rate  $da/dt = R(a)$ ,  $a$  being crack length and  $t$  time, while  $R$  is a single-valued function.

In practice  $R$  is random, as in the first trajectory of Figure 1. Each growth increment during cycling is random and there is also an overall materials effect acting for every load applied to a given structure. The first of these two types of randomness corresponds to some stochastic differential or difference equation and leads to the second order diffusion term in the Fokker-Planck equation. The second overall type of variability allows smooth crack growth for any particular structure but variations between them as shown on the right of the figure.

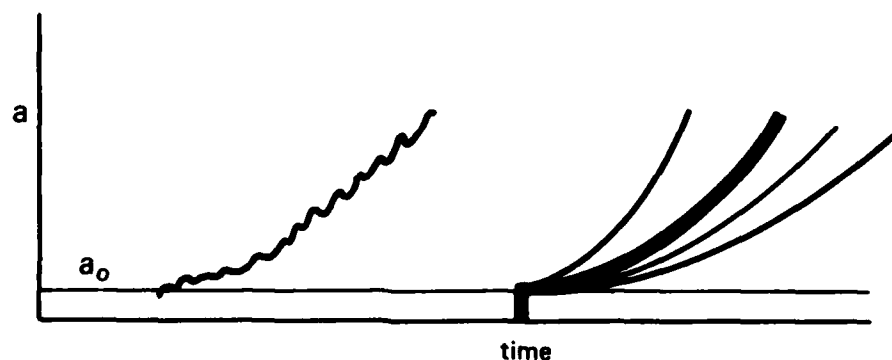


FIG. 1 NATURE OF CRACK GROWTH



If diffusion is ignored but overall variations are allowed for, the fatigue model—though different—is still deterministic and follows the previous continuity equation

$$\begin{aligned} \frac{Df}{Dt} &= \frac{\partial f}{\partial t} + R(a, t) \frac{\partial f}{\partial a} \\ &= -f(\text{div } R(a, t) + r(a)) \end{aligned} \quad (2.1)$$

for crack length density  $f(a, t)$  given time  $t$ .

## 2.2 Rate Divergence

We now assume that for a crack beginning at  $t_0$ ,  $da/dt = XR_0(a)$  which replaced  $R(a, t)$  above. Here  $X$  determines the overall crack rate and is a function of  $a$ ,  $t$  and  $t_0$ . Without knowledge of  $a/t$ ,  $X$  is random with density  $f_X(x)$ , log-Normal where necessary; the particular trajectory corresponding to any  $x$  is a transformation of  $X$  to  $A, t$ .

Then the divergence

$$\frac{\partial R}{\partial a} = x \frac{dR_0}{da} + R_0(a) \frac{\partial x}{\partial a}$$

with  $t$  and  $t_0$  fixed.

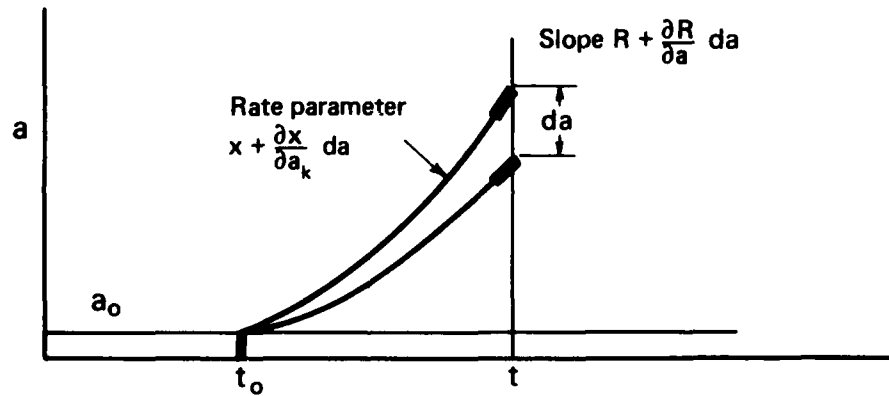


FIG. 2 RATE DIVERGENCE

For any crack  $da/dt$  integrates to

$$X = \frac{1}{t - t_0} \int_{a_0}^a \frac{da}{R_0(a)}$$

whence

$$\frac{\partial X}{\partial a} = \frac{1}{(t - t_0) R_0(a)} \quad (2.2)$$

and

$$\frac{\partial R}{\partial a} = X \frac{dR_0}{da} + \frac{1}{t - t_0} \quad (2.3)$$

This is illustrated by Figure 2.

### 2.3 Crack Length Density

With the present model, the characteristic equations of (2.1) are

$$\frac{-df}{f(xR_0'(a)+r(a)+1/(t-t_0))} = dt = \frac{da}{xR_0(a)}. \quad (2.4)$$

Along the crack trajectories (for which  $x$  is constant)

$$\begin{aligned} -d \log f &= \frac{R_0'(a)}{R_0(a)} da + \frac{r(a)}{xR_0(a)} da + \frac{da}{x(t-t_0)R_0(a)} \\ &= d \log R_0(a) + \frac{r(a)}{xR_0(a)} da + \frac{da}{t_a R_0(a)}. \end{aligned}$$

Integrating,

$$\log f = A - \log R_0(a) - \frac{1}{x} \int_{a_0}^a \frac{r(a)}{R_0(a)} da - \int_{a_0}^a \frac{da}{t_a R_0(a)}.$$

Now let  $t_a$  be the growth time of the median crack ( $x=1$ ) and change the crack length variable  $a$  to  $t_a$ , the time for the median crack to reach the same size.

Then

$$\log f = A - \log R_0(a) - \frac{1}{x} \int_0^{t_a} r(c) dc - \log t_a. \quad (2.5)$$

This is singular at initiation  $t_a = 0$  because all cracks here begin at the one point  $(t_0, a_0)$  and (2.5) indicates an infinite density corresponding to the concentrated probability that  $a|t_0 = a_0$ . However, the equation suggests that we express  $f$  in terms of a modified "density"  $ft_a$ . Thus,

$$f(a|t)t_a = \frac{A}{R_0(t_a)} \exp\left(-\frac{1}{x} \int_0^{t_a} r(c) dc\right). \quad (2.6)$$

where  $A$  is constant along each trajectory, as is  $x$ . Consider another median growth time  $x\tau$  for which

$$f(x\tau) = \frac{A}{R_0(x\tau)} \exp\left(-\frac{1}{x} \int_0^{x\tau} r(c) dc\right).$$

Substituting for  $A$  in (2.6) leads to

$$f(a|t)t_a = R_0(x\tau) \frac{f(a|\tau, x)x\tau}{R_0(t_a)} \exp\left(-\frac{1}{x} \int_{x\tau}^{t_a} r(c) dc\right). \quad (2.7)$$

Here, of course,  $t_a = x(t-t_0)$  and since  $x$  is constant along a characteristic, it may be cancelled from each side to give

$$f \cdot (t-t_0) = \tau f(a|\tau, x) \frac{R_0}{R_0(x(t-t_0))} \exp\left(-\frac{1}{x} \int_0^{x_t} r(c) dc\right). \quad (2.8)$$

where  $R_0 = R_0(0)$ , the initial growth rate.

#### 2.3.1 Boundary Conditions

These will be expressed in terms of  $f(a|t)t_a$ . Being supported only at  $(t_0, a_0)$ , they must be regarded asymptotically. So far we have not used the assumed density  $f_X(x)$ , nor any assumptions about asymptotic behaviour.

As in Figure 3 we assume a finite initial crack rate (unless  $x = 0$ ) already implied by our use of  $R_0 = R_0(0)$ . Together with continuity, this presumes asymptotically linear initial growth consistent with the two-stage fatigue assumption of a fixed finite initial crack length. In welded structures, where immediate initiation is more plausible, the same behaviour agrees with the assumption of a pre-existing flaw, restricted, however, to a constant size.

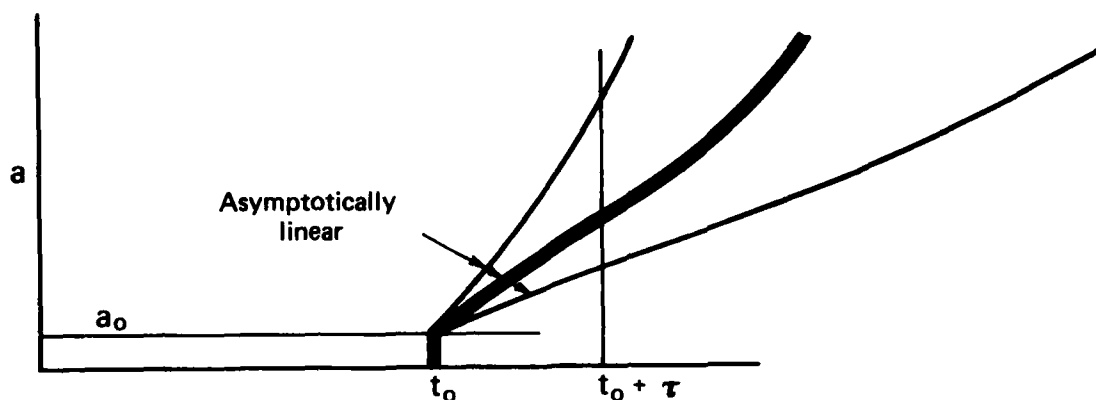


FIG. 3 APPROACH TO BOUNDARY CONDITIONS

As  $\tau \rightarrow 0$

$$a \sim x R_0 \tau$$

and the asymptotic crack length density

$$f(a, \tau) = f_X(x) / R_0 \tau,$$

so that

$$\tau f(a, \tau) \sim f_X(x) / R_0$$

and (2.8) becomes

$$\begin{aligned} f(a, t, t_0) &= \frac{f_X(x)}{(t - t_0) R_0(x(t - t_0))} \exp \left( - \frac{1}{x} \int_0^{x_t} r(c) dc \right) \\ &= \frac{x f_X(x)}{t_a R_0(t_a)} \exp \left( - \frac{1}{x} \int_0^{t_a} r(c) dc \right) \end{aligned} \quad (2.9)$$

## 2.4 Density of Crack Life

At a given time, the total rate of attrition is

$$\phi(t, t_0) = \int_{a_0}^{\infty} \frac{f_X(x)}{(t - t_0) R_0(t_a)} r(t_a) \exp \left( - \frac{1}{x} \int_0^{t_a} r(c) dc \right) da \quad (2.10)$$

remembering that median growth for time  $t_a = x(t - t_0)$  produces the crack  $(a, t)$ . Again,  $t_a$  is the natural variable of integration and we obtain

$$\phi(t|t_0) = \int_0^\infty \frac{f_X(t_a/(t-t_0))}{t-t_0} r(t_a) \exp\left(-\frac{t-t_0}{t_a} \int_0^{t_a} r dc\right) dt_a \quad (2.11)$$

When cracks are small one expects this to be consistent with attrition at the constant risk  $r_0$ ; we show that this is  $r_0$ .

Near  $t = t_0$ , the reliability type density from (2.11)

$$r(t_a) \exp(\dots) \rightarrow r_0 \exp(-r_0(t-t_0)) \quad (2.12)$$

for all cracks from  $(t_0, a_0)$ , i.e. all values of  $x$ . This factor, the desired asymptotic behaviour, may be taken outside the integration. The remaining integral is asymptotically one since

$$f_X(t_a/(t-t_0)) = f_X(x),$$

a density function.

### 3. RUNAWAY CRACKS

These will be treated by telescoping as in Reference 1. The present assumption is that the median crack "runs away" when  $t_a = T_a$  so that other cracks are runaways after the actual times  $T_a/X$ . Mathematically this means that  $r(T_a+) \rightarrow \infty$  so that the risk integrated past  $T_a$  is also infinite.

#### 3.1 Reduction of Life Range

Equation (2.11) may then be written

$$\int_0^{T_a-} + \int_{T_a-}^{T_a+} + \int_{T_a+}^\infty \frac{f_X(x)}{t-t_0} r(t_a) \exp\left(-\frac{1}{x} \int_0^{t_a} r dc\right) dt_a \quad (3.1)$$

and we first consider the third integral in the limit as risk uniformly increases.

Thus, let  $R > r(t_a) > \rho$  when  $t_a > T_a$  so that

$$R \cdot (t-t_0) > \frac{t-t_0}{t_a} \int_0^{t_a} r(c) dc > \rho(t-t_0) \quad (x = t_a/(t-t_0))$$

in which case

$$e^{-R \cdot (t-t_0)} < \text{exponential term} < e^{-\rho(t-t_0)}$$

Let  $f_X(x)/(t-t_0)$  be bounded. We require an upper bound for

$$\begin{aligned} \int_{T_a+}^\infty \frac{f_X r e^{-\int r dc} x}{t-t_0} dt_a &< \int_{T_a+}^\infty f_X \left( \frac{t_a}{t-t_0} \right) R e^{-\rho(t-t_0)} \frac{dt_a}{t-t_0} \\ &< R e^{-\rho(t-t_0)} \{1 - F_X(T_a/(t-t_0))\}. \end{aligned}$$

If now there are finite constants  $C, d$  such that  $R < C\rho^d$  then

$$\begin{aligned} R e^{-\rho(t-t_0)} &< C \rho^d e^{-\rho(t-t_0)} \\ &\rightarrow 0 \quad \text{as } \rho \rightarrow \infty \end{aligned}$$

Thus if  $C\rho^d > r(t_a|t_a > T_a) > \rho$  the third integral in (3.1) vanishes as  $r(T_a+) \rightarrow \infty$ .

#### 3.2 Telescoping

Now consider the second term

$$\frac{1}{t-t_0} \int_{T_a-}^{T_a+} f_X \left( \frac{t_a}{t-t_0} \right) r(t_a) \exp\left(-\frac{t-t_0}{t_a} \int_0^{t_a} r(c) dc\right) dt_a$$

$$= \frac{f_X(T_a/(t-t_0))}{t-t_0} \int_{T_a}^{T_a} r(t_a) \exp \left( -\frac{1}{x} \int_0^{t_a} rdc \right) dt_a + O(\Delta T), \quad (3.2)$$

using the mean value theorem.

Apart from the factor  $1/x$  the integrand above is a probability density of life appropriate to a risk  $r(c)$ . To develop this idea consider a particular crack trajectory, along which  $x$  is constant. Then reliability theory indicates the conditional life distribution

$$\begin{aligned} \Phi_T(t|t_0, x) &= 1 - \exp \left( - \int_{t_0}^t r(x(t'-t_0)) dt' \right) \\ &= 1 - \exp \left( - \frac{1}{x} \int_0^{t_a} r(c) dc \right) \end{aligned} \quad (3.3)$$

with  $c = x(t'-t_0)$  and  $t_a = x(t-t_0)$ . This has the density derivative

$$\phi_T(t|t_0, x) = x r(t_a) \exp \left( - \frac{1}{x} \int_0^{t_a} r(c) dc \right), \quad (3.4)$$

the integrand of (3.2) apart from  $x$ , here equal to  $(t-t_0)/T_a + O(\Delta T)$ . Thus (3.3) provides the integral in (3.2) and with runaway cracks (3.3) is one if  $t_a = T_a$ . With all substitutions the limit of (3.1) is

$$\begin{aligned} \phi(t|t_0) &= \int_0^{T_a} \frac{f_X(x)}{t-t_0} r(t_a) \exp \left( - \frac{1}{x} \int_0^{t_a} r(c) dc \right) dt_a \\ &= \frac{1}{T_a} f_X \left( \frac{T_a}{t-t_0} \right) \exp \left( - \frac{t-t_0}{T_a} \int_0^{T_a} r(c) dc \right). \end{aligned} \quad (3.5)$$

In this generalisation of (2.11)  $T_a$ , the median time to runaway cracking, is effectively a constant parameter. The second term, the density of runaways, is similar to but not the same as the corresponding term with deterministic cracking; the same idea of concentrated crack life attrition still applies.

#### 4. LIFE DISTRIBUTION

From Reference 1, the life distribution for translated cracks

$$\begin{aligned} \phi(t) &= r_0 e^{-r_0 t} (1 - F_0(t)) + \int_0^{T_a} e^{-r_0 t} f_0(t_0) dF_a(t-t_0) \\ &= e^{-r_0(t-T_a)} f_0(t-T_a) \exp \left( - \int_0^{T_a} r(c) dc \right), \end{aligned} \quad (4.1)$$

where if  $t_a = t - t_0$

$$dF_a/dt_a = r(t_a) \exp \left( - \int_0^{t_a} r(c) dc \right) \quad (4.1A)$$

from reliability theory applied to crack life.

The moment generating function here is

$$M_\phi(u) = M_t(u - r_0)M_a(u) + \frac{r_0}{u - r_0} (M_t(u - r_0) - 1) \quad (4.2)$$

where  $M_t(u)$  and  $M_a(u)$  generate moments of initial life and crack time.

If  $M_a$  is replaced by another function reflecting the outcome of attrition on randomly cracking structures from  $t = 0$ , the unique inversion of (4.2) indicates that  $\phi(t)$  is similarly generalised by generalising the reliability density (4.1A), the inverse of the new  $M_a(u)$ .

For the different but deterministic crack model above, this replacement density is (3.5), which already includes any runaway cracks so that the third term in (4.1) is subsumed into the second.

For completeness, the generalised life distribution will be found directly. For this, one needs  $f(a't, t_0)$  which is (2.9) in the present instance. The formula (3.5) is also the density of  $t_A = t - t_0$ , the crack part of the total life. Equation (3.5) then indicates that  $t_A$  is independent of initiation, as one would expect, and the total rate of attrition is thus

$$\phi(t) = r_0 e^{-r_0 t} (1 - F_0(t)) + \int_0^t e^{-r_0 t_0} f_0(t_0) \phi_A(t_A) dt_0, \quad t_0 = t - t_A, \quad (4.3)$$

where from (3.5)

$$\begin{aligned} \phi_A(t_A) &= \int_0^{T_a/t_A} f_X(x) r(x t_A) \exp \left( -\frac{1}{x} \int_0^{x t_A} r(c) dc \right) dx \\ &= \frac{1}{T_a} f_X \left( \frac{T_a}{t_A} \right) \exp \left( -\frac{t_A}{T_a} \int_0^{T_a} r(c) dc \right). \end{aligned} \quad (4.4)$$

Because renewals affect only the initial life,<sup>5</sup> the form (4.2) may be immediately applied when there are inspections or renewals.

#### 4.1 Moment Generating Function of Crack Time

This is the Laplace transform of (4.4) which in general is difficult to simplify. To investigate the effect of small departures from translatable cracking, we shall now assume that  $X$  is log-Normal. Then, for  $x$  near 1, we shall try to relate the extended MGF  $M_A$  to that of the median crack  $M_a$ .

From the Laplace transform of (4.4)

$$\begin{aligned} M_A(u) &= \int_0^\infty \int_0^{T_a/t_A} f_X(x) r(t_a) \exp \left( -\frac{1}{x} \left[ u t_a + \int_0^{t_a} r dc \right] \right) dx dt_A \\ &= \int_0^\infty \frac{1}{T_a} f_X \left( \frac{T_a}{t_A} \right) \exp \left( -\frac{t_A}{T_a} \left[ u T_a + \int_0^{T_a} r dc \right] \right) dt_A \end{aligned}$$

In both terms put  $x = e^\xi$ ,  $t_A = t_a e^{-\xi}$  with the Jacobian  $J = 1$ . In the first term, the upper limit of  $x = T_a/t_A$  from which we need  $\xi$ . On the right, we have

$$x = T_a e^\xi / t_a \quad \text{leaving} \quad t_a \equiv T_a$$

with this upper limit for any value of  $\xi$ . Thus, the range of  $\xi$  is  $(-\infty, \infty)$  whilst  $0 < t_a < T_a$ .

In the second term,  $x \equiv T_a/t_A$  for all  $t_A$ , and with this we find  $t_a = t_A x \equiv T_a$ ; that is, the range of  $t_a$  is just the point  $T_a$ . On the other hand, this identity and the fact that  $0 < t_A < \infty$  means that  $x$  has the same range. Thus, in the runaway term, the two substitutions have changed an integration over  $t_A$  to an averaging over  $\xi$ . It would be possible to avoid some of this manip-

ulation if the MGF were originally defined as a Stieltjes integral with a concentrated probability of runaway cracking.

With these changes

$$M_A(u) = \int_0^{T_a} \int_{-\infty}^{\infty} r(t_a) e^{-e^{-\xi G(t_a)}} dF_{\xi}(\xi) dt_a \\ = \int_{-\infty}^{\infty} e^{-e^{-\xi G(T_a)}} dF_{\xi}(\xi), \quad (4.5)$$

where

$$G(t_a) = ut_a + \int_0^{t_a} r(c) dc.$$

#### 4.2 Crack Life MGF for Small Variance

Now assume that  $\xi \sim N(0, \sigma^2)$  and expand about  $\xi = 0$ . In this section and the next steps to successively cruder but more practical approximations are marked by equations. If we use

$$G(t_a) = ut_a + H(t_a)$$

and take three terms of  $G \exp - \xi$ , then

$$M_A(u) \approx \int_0^{T_a} r(t_a) e^{-ut_a - H(t_a)} \exp \left( \frac{\sigma^2 G^2/2}{1 + \sigma^2 G} \right) (1 + \sigma^2 G)^{-1/2} dt_a \\ + e^{-G(T_a)} \exp \left( \frac{\sigma^2 G/2}{1 + \sigma^2 G} \right) (1 + \sigma^2 G(T_a))^{-1/2}. \quad (4.6)$$

It is now convenient to subsume these two terms into one Stieltjes integral over a semi-infinite range with the probability density  $dF_a(t_a)$ . The terms in (4.6) containing variance expand as

$$\exp \left( \frac{\sigma^2 G^2/2}{1 + \sigma^2 G} \right) (1 + \sigma^2 G)^{-1/2} = 1 + \frac{1}{2} \sigma^2 (H^2 - H) - \frac{1}{4} \sigma^4 (\frac{3}{2} H^2 - H^3 + \frac{1}{2} H^4) + \dots \\ ut_a \{ 1 + \frac{1}{2} \sigma^2 (H^2 - 3H + 1) - \frac{1}{4} \sigma^4 (-3H + 4\frac{1}{2} H^2 - 3H^3 + \frac{1}{2} H^4) + \dots \} \\ + \frac{1}{2} u^2 t_a^2 \{ 1 + \frac{1}{2} \sigma^2 (H^2 - 3H + 2) - \frac{1}{4} \sigma^4 (3 - \frac{3}{2} H^2 + 3H^3 - \frac{1}{2} H^4) + \dots \} \\ 0(u^3 t_a^3 \text{ etc}). \quad (4.7)$$

Here the sum of expectations of the first term on each line obviously form  $M_a(u)$ , generating moments of the median crack life. If  $u = 0$ , the kernel corresponding to the first line would (if complete) be the probability density (3.5) of our present model. For  $M_A$ , the expectation, however, is taken over median crack lives; the first term is one, so that the variance terms on the first line have zero expectations. In (4.7) therefore, multiples of the variance terms of the first line may be subtracted from the others within an accuracy  $O(\sigma^6)$ . This simplifies the approximation to

$$M_A(u) \approx M_a(u) - u \int_0^{\infty} t_a \{ \frac{1}{2} \sigma^2 (1 - 2H) - \frac{1}{4} \sigma^4 (\frac{3}{2} (H^2 - H) - H^3) \} dF_a(t_a) \\ + \frac{1}{2} u^2 \int_0^{\infty} t_a^2 \{ \sigma^2 (1 - H) - \frac{1}{4} \sigma^4 (\frac{3}{4} - H^3) \} dF_a(t_a) \quad (4.8)$$

where the MGF of the median life

$$M_a(u) = \int_0^{\infty} e^{-ut} dF_a(t).$$

Obviously, the  $u, u^2$  terms in (4.8) represent increases in moments caused by the Normal scatter of crack rates. Similarly, the first line of (4.7) can be interpreted as the corresponding change in crack life density. Although all the integrals converge the errors introduced in (4.6) by the quadratic approximation for  $\exp -\xi$  will be large unless  $\sigma < 1$  which is true for fatigue.

### 4.3 Heuristic Approximations of Crack Life Moments

In this section, we derive a rule of thumb to estimate the effect of crack rate variability. To proceed further from (4.7) or (4.8) it is necessary to avoid higher moments of  $F_a$ . These are therefore replaced by the corresponding cumulants.

We could also bound the risk as

$$H(t_a) \leq g t_a / \mu_a = h t_a, \text{ say,} \quad (4.9)$$

and find strict bounds for the moments. This amounts to approximating the given crack-life risk problem by a constant risk problem. Since these bounds are likely to be very conservative, we shall abandon rigour henceforth and regard  $g = O(1)$  as an averaging correction, retaining all signs. This will provide heuristic estimates of changes and indicate their possible importance.

Treating (4.9) as a simple substitution

$$M_A(u) \sim M_a(u) - u \left\{ \frac{1}{2} \sigma^2 \mu_a - \sigma^2 h (1 - \frac{3}{4} \sigma^2) \mu_{2a} - \frac{3}{4} h^2 \sigma^4 \mu_{3a} + \frac{1}{2} h^3 \sigma^4 \mu_{4a} \right\} \\ + \frac{1}{2} u^2 \left\{ \sigma^2 (1 - \frac{3}{4} \sigma^2) \mu_{2a} - h \sigma^2 \mu_{3a} - h^3 \sigma^4 \mu_{5a} \right\}.$$

Now substitute cumulant formulae for the moments  $\mu_{ra}$  with a view to then neglecting those above  $\kappa_2$  (we retain the notation  $\mu_a, \sigma_a^2$  for the first two cumulants). This is done because cumulants, unlike moments, can decrease with order. Thus, neglecting higher cumulants and terms in  $\sigma^6$  and higher, we find

$$M_A(u) \sim M_a(u) \\ - u \left\{ \mu_a \left[ \frac{1}{2} \sigma^2 - g \sigma^2 (1 - \frac{3}{4} \sigma^2 (1 - g)) \right] + \frac{1}{2} g^3 \sigma^4 \right\} + (\sigma_a^2 / \mu_a) \left\{ - g \sigma^2 (1 - \frac{3}{4} \sigma^2) - 2 \cdot 25 g^2 \sigma^4 + 3 g^3 \sigma^4 \right\} \\ + \frac{3}{2} g^3 \sigma^4 \sigma_a (\sigma_a / \mu_a)^3 \} \\ + \frac{1}{2} u^2 \left\{ (\mu_a^2 + \sigma_a^2) [\sigma^2 (1 - \frac{3}{4} \sigma^2) - g \sigma^2 - g^3 \sigma^4] - \sigma_a^2 [2 g \sigma^2 + 9 g^3 \sigma^4 + 15 g^3 \sigma^4 (\sigma_a / \mu_a)^2] \right\} \quad (4.10)$$

where  $g \approx 1$ . If we now make all the  $g$ s one, this eventually leads to

$$\mu_A / \mu_a \sim 1 + 0.6 \sigma^2 (1 + \sigma^2) - (\sigma_a / \mu_a)^2 [1 - 1.5 \sigma^2 (1 + (\sigma_a / \mu_a)^2)] \\ (\sigma_A / \sigma_a)^2 \sim 1 - \sigma^2 (5 - \sigma^2 [2.25 + (\sigma_a / \mu_a)^2 \{14 - 3(\sigma_a / \mu_a)^2\}]) \quad (4.11)$$

where  $\sigma^2$  is the logarithmic variance of crack growth rates, while  $\mu_a$  and  $\sigma_a$  refer to reliability type crack lives based on the median crack growth curve.

#### 4.3.1 Example

Equation (4.11) has been derived from reliability based lives for structures which crack immediately. For the general two-stage model of fatigue it indicates the effect of random rates on cracking times.

The model of crack rates is the coarse randomness in which the possible crack trajectories are randomly scaled versions of the same median curve. As a typical example, put

$$\sigma = 0.2303, \quad \sigma_a / \mu_a = (e^{\sigma^2} - 1) = 0.2589$$

which corresponds to a log-Normal life, with logarithmic standard deviation<sup>6</sup> 0.1 (to base 10). To base 10 the crack rate standard deviation<sup>7</sup> is also 0.1; the quantities in (4.11) refer to natural logarithms.



After substitution, we find  $\mu_A/\mu_a = 1 - 0.03344$ ,  $(\sigma_A/\sigma_a)^2 = 1 - 0.25617$  and  $\sigma_A/\sigma_a = 1 - 0.13754$ .

The general effect indicated seem to be a small reduction in mean crack time and a possibly more important reduction of its variance.

## 5. CONCLUSIONS

The methods of Reference 1 have been applied to one-crack models in which all the crack trajectories are smooth, but form a family in which the growth rate for a given crack length is a random factor of the median rate. The application is straightforward and it is simple to allow for runaway cracks. There is no simple relation to the results for median crack lives, although asymptotic effects on the moments are suggested.

Application to a typical example seems to indicate that the mean life with random cracking is slightly less than that from the median crack, but there is a more important reduction of variance. These results are being sharpened by further study of the asymptotic moment generating function.

The model described above refers to structures in which cracking begins immediately. In Section 4 the generalised density corresponding to this has been used to extend the previous<sup>1</sup> two-stage model. This allows randomness in both the initiation and crack growth stages.

## REFERENCES

1. Ford, D. G. Reliability and Structural Fatigue in One-Crack Models. ARL Structures Report 369, July 1978.
2. Ford, D. G. The Development of the Theory of Structural Fatigue. Aircraft Structural Fatigue —Proceedings of a Symposium held in Melbourne, 19-20 October 1976. ARL Structures Report 363, April 1977.
3. Diamond, Patricia, and Payne, A. O. Reliability Analysis Applied to Structural Tests. Proceedings of Symposium on Advanced Approaches to Fatigue Evaluation, Miami Beach, May 1971; NASA SP-309, 1972.
4. Hooke, F. H. A New Look at Structural Reliability and Risk Theory. Presented at AIAA/ASME 14th Structures, Structural Dynamics and Materials Conference, Bethesda, Md., 3-5 April 1978. (To appear in AIAA Journal.)
5. Ford, D. G. Structural Fatigue in One-Crack Models with Arbitrary Inspection. ARL Structures Report 377, April 1979.
6. Jost, G. S., and Verinder, F. E. A Survey of Fatigue Life Variability in Aluminium Alloy Aircraft Structures. ARL Structures Report 329, February 1971.
7. Heath, W. G., Nicholls, L. F., and Kirkby, W. T. Practical Application of Fracture Mechanics Techniques to Aircraft Structural Problems. "Fracture Mechanics Design Methodology", AGARD Conference Proceedings No. 221, February 1977.

## DISTRIBUTION

Copy No.

### AUSTRALIA

#### Department of Defence

##### Central Office

Chief Defence Scientist	1
Deputy Chief Defence Scientist	2
Superintendent, Science and Technology Programs	3
Defence Library	4
Document Exchange Centre, DISB	5 21
Joint Intelligence Organisation	22
Australian Defence Science and Technical Representative (UK)	
Counsellor, Defence Science (USA)	

##### Aeronautical Research Laboratories

Chief Superintendent	23
Library	24
Superintendent Structures Division	25
Divisional File Structures	26
Author: D. G. Ford	27 28
G. S. Jost	29
A. D. Graham	30
J. Q. Clayton	31

##### Materials Research Laboratories

Library	32
---------	----

##### Defence Research Centre, Salisbury

Library	33
---------	----

##### Central Studies Establishment

Information Centre	34
--------------------	----

##### Engineering Development Establishment

Library	35
---------	----

##### RAN Research Laboratory

Library	36
---------	----

##### Defence Regional Office

Library	37
---------	----

##### Navy Office

Naval Scientific Adviser	38
--------------------------	----

##### Army Office

Army Scientific Adviser	39
Royal Military College, Library	40

##### Air Force Office

Aircraft Research and Development Unit, Scientific Flight Group	41
Air Force Scientific Adviser	42
Technical Division Library	43

D. Air Eng. AF	44	
HQ Support Command (SENGSO)	45	
RAAF Academy, Point Cook	46	
<b>Department of Productivity</b>		
<b>Government Aircraft Factories</b>		
Library	47	
Mr. L. Tuller	48	
<b>Department of Transport</b>		
Secretary	49	
Library	50	
Airworthiness Group: Mr. K. O'Brien	51	
Mr. C. Torkington	52	
Accident Group (Mr. I. S. Milligan)	53	
<b>Statutory, State Authorities and Industry</b>		
CSIRO Mechanical Engineering Division, Chief	54	
CSIRO Material Science Division, Director	55	
Qantas, Library	56	
Trans-Australia Airlines, Library	57	
Ansett Airlines, Library	58	
BHP Melbourne Research Laboratories	59	
Commonwealth Aircraft Corporation, Manager	60	
Commonwealth Aircraft Corporation, Manager of Engineering	61	
Commonwealth Aircraft Corporation, Mr. R. C. Beckett	62	
Commonwealth Aircraft Corporation, Mr. H. Granow	63	
Hawker de Havilland Pty. Ltd., Librarian, Bankstown	64	
<b>Universities and Colleges</b>		
Adelaide	Barr Smith Library	65
Deakin	Library	66
	Dr. B. Bathgate	67
Flinders	Library	68
James Cook	Library	69
La Trobe	Library	70
Melbourne	Engineering Library	71
	Engineering Dr. J. Williams	72
Monash	Library	73
	Prof. I. J. Polmear, Materials Engineering	74
Newcastle	Library	75
New England	Library	76
New South Wales	Prof. Traill-Nash, Structural Engineering	77
Sydney	Engineering Library	78
	Prof. G. A. Bird, Aeronautical Engineering	79
Queensland	Engineering Library	80
Tasmania	Engineering Library	81
	Prof. R. A. Oliver, Civil and Mechanical Engineering	82
Western Australia	Library	83
RMIT	Library	84
	Mr. H. Millicer, Aeronautical Engineering	85
	Mr. Pugh, Mechanical Engineering	86
<b>CANADA</b>		
Aluminium Labs. Ltd., Library		87
CAARC Co-ordinator (Structures)		88
NRC, National Aeronautical Establishment, Library		89
NRC, National Aeronautical Establishment, Div. of Mech. Eng. (Dr. D. McPhail)		90

<b>Universities</b>		
McGill	Library	91
Toronto	Institute of Aerophysics	92
Waterloo	Dept. of Mechanical Engineering	93
	Dept. of Civil Engineering	94
<b>FRANCE</b>		
ONERA, Library		95
Service de Documentation, Technique de l'Aeronautique		96
<b>GERMANY</b>		
LBF, Dr. D. Schütz		97
DVLK, Dr. W. Schütz		98
<b>INTERNATIONAL COMMITTEE ON AERONAUTICAL FATIGUE</b>		
per Australian ICAF Representative		99-122
<b>INDIA</b>		
CAARC Co-ordinator (Structures)		123
Civil Aviation Department, Director		124
Defence Ministry, Aero. Development Est., Library		125
Indian Institute of Science, Library		126
Indian Institute of Technology, Library		127
National Aeronautical Laboratory, Director		128
<b>ITALY</b>		
Associazione Italiana di Aeronautica e Astronautica		129
Fiat Co., Dr. G. Gabrielli		130
Universita degli Studi di Pisa, Dr A. Salvetti		131
<b>ISRAEL</b>		
Technion Israel Institute of Technology, Prof. J. Singer		132
Technion Israel Institute of Technology, Professor A. Buch		133
<b>JAPAN</b>		
National Aerospace Laboratory, Library		134
<b>Universities</b>		
Tokyo	Inst. of Space and Aerospace, Library	135
Tohoku (Sendai)	Library	136
<b>Research Institute of Strength and Fracture of Materials</b>		
Kobe, Professor T. Nakagawa		137
Kyushu Institute of Technology, Library		138
<b>NETHERLANDS</b>		
Central Org. for Applied Science Research TNO, Library		139
National Aerospace Laboratory (NLR), Library		140
Delft University of Technology		141
<b>NEW ZEALAND</b>		
Defence Science Establishment, Librarian		142
Transport Ministry, Civil Aviation Division Library		143
<b>Universities</b>		
Canterbury	Library	144

## SWEDEN

Aeronautical Research Institute	145
Chalmers Institute of Technology, Library	146
Kungliga Tekniska Hogskolan	147
SAAB-Scania, Library	148

## SWITZERLAND

F. & W. Ltd.	149
--------------	-----

## UNITED KINGDOM

CAARC, Secretary	150
Aeronautical Research Council, Secretary	151
Royal Aircraft Establishment:	
Farnborough, Library	152
Bedford, Library	153
Commonwealth Air Transport Council Secretariat	154
British Library, Science Reference Library	155
Lending Division	156
Fulmer Research Institute Ltd., Research Director	157
Science Museum Library	158
Welding Institute Library	159
National Engineering Laboratories, Superintendent	160
Naval Construction Research Establishment, Superintendent	161
CAARC Co-ordinator, Structures	162
Aircraft Research Association, Library	163
British Ship Research Association	164
Central Electricity Generating Board	165
Motor Industries Research Association, Director	166
Rolls-Royce Aeronautics Division, Library	167
Rolls-Royce Bristol Siddeley Division, Library	168
British Aerospace Corporation:	
Aircraft Group, Kingston-Brough Division	169
Aircraft Group, Headquarters Library, Kingston-upon-Thames	170
Aircraft Group, Manchester Division	171
Aircraft Group, Weybridge-Bristol Division	172
Aircraft Group, Warton Division	173
British Hovercraft Corporation Ltd., Library	174
Short Brothers & Harland	175
Westland Helicopters Ltd.	176

### Universities and Colleges

Bristol	Library, Engineering Department	177
Cambridge	Library, Engineering Department	178
London	Library, Department of Aeronautics	179
Manchester	Library, Engineering	180
Nottingham	Library	181
Southampton	Library	182
Strathclyde	Library	183
Cranfield Institute		
of Technology	Library	184
Imperial College	The Head	185
	Professor of Mechanical Engineering	186
	Library, Department of Aeronautics	187

## UNITED STATES OF AMERICA

NASA Scientific and Technical Information Facility	188
Sandia Group Research Organisation	189

American Institute of Aeronautics and Astronautics	190
Applied Mechanics Reviews	191
Boeing Company, Head Office, Mr Watson	192
The John Crerar Library	193
US Atomic Energy Commission, Division of Tech. Information	194
Cessna Aircraft Co., Executive Engineer	195
Battelle Memorial Institute, Library	196
Dr D. Broe k	197
General Dynamics	198
General Electric, R & D Manager, Materials Science & Engineering	199
Lockheed Georgia Company	200
MacDonnel-Douglas	201
Metals Abstracts	202
Air Force Flight Dynamics Labs., Wright-Patterson AFB	203
NASA Langley (Virginia), John H. Crews Jnr.	204
J. R. Davidson	205
National Bureau of Standards, Library	206
Library of Congress, Gift and Exchange Dept.	207
US Dept. of Transportation, Transportation Systems Center,	208
Dr Elaine I. Savage	
US Naval Research Laboratory, Washington D.C.	209
US Naval Air Development Center	210

#### Universities and Colleges

Brown	Professor R. E. Meyer	211
California	Dept. of Aerosciences, Dr M. Holt	212
Cornell (New York)	Library, Aeronautics Laboratories	213
Florida	Dept. Aeronautical Engineering, M. H. Clarkson	214
Harvard	Library, Engineering	215
Illinois	Dept. of Theoretical and Applied Mechanics	216
Iowa State	Mechanical Engineering, Dr G. K. Sekory	217
Johns Hopkins	Dept. of Mechanical Engineering	218
Massachusetts	Library, School of Engineering	219
Pennsylvania	Library, College of Engineering	220
Stanford	Library, Dept. of Aeronautics	221
George Washington	Professor J. N. Yang	222
Wisconsin	Memorial Library, Serials Dept.	223
California Institute of Technology	Library, Gradvale Aeronautical Labs.	224
Massachusetts Institute of Technology	Library	225
Polytechnic Institute of New York	Aeronautical Labs., Library	226

Spares

227-245

**DAT**  
**ILM**